

Precise Robust Inertial Guidance for Munitions: Navigation-Grade Inertial Measurement Unit (PRIGM:NGIMU) Proposers' Day

Dr. Robert Lutwak, Program Manager
DARPA Microsystems Technology Office

April 17, 2015
Arlington, VA





Proposers' Day Agenda

- PRIGM:NGIMU Introduction and BAA Overview
 - Program objectives
 - BAA and proposal process
 - Government T & E capabilities

- Open Q & A Session
 - Write your questions on the notecards provided
 - Submit questions before the Q & A session begins

- Attendee Presentations
 - 12 minutes to speak, 3 minutes for questions
 - Please help maintain the schedule

Nav

GBU-53 SDB-II



© Wikimedia Commons

GBU-31 JDAM

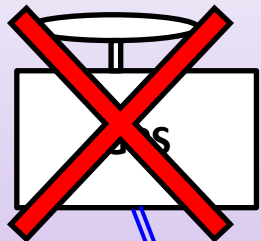


Source: cat-uxo.com

AGM-114 Hellfire

Source: defense-update.com

Navigation System



INS

Algorithms

Output:

 x, y, z, v_x, v_y, v_z

Inertial Navigation System (INS)

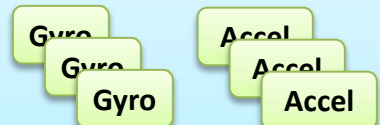
Clock

IMU

Output:

 x, y, z, v_x, v_y, v_z

Inertial Measurement Unit (IMU)



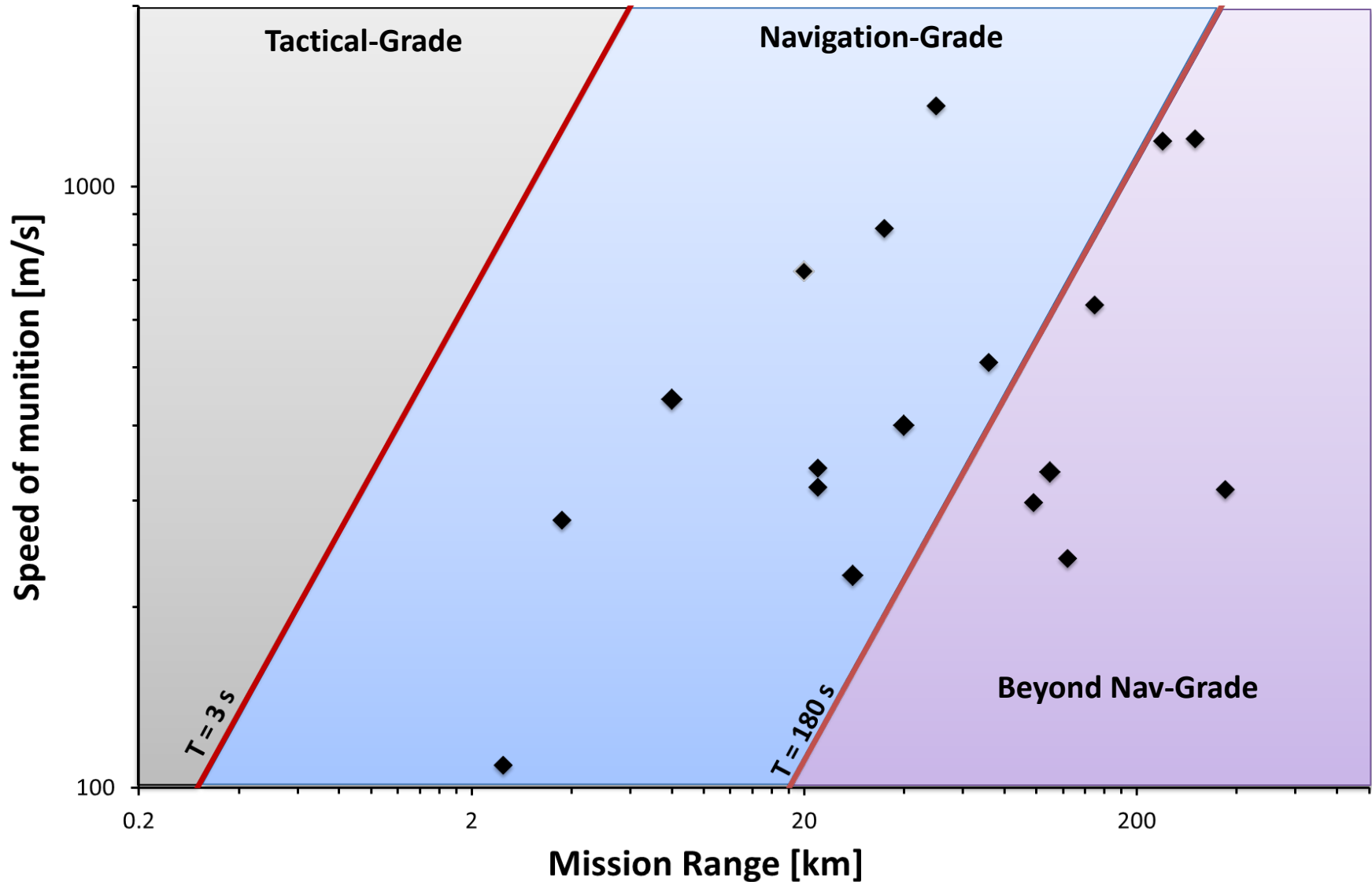
Sensor Control & Readout

Output:

 $a_x, a_y, a_z, \dot{\alpha}, \dot{\beta}, \dot{\gamma}$



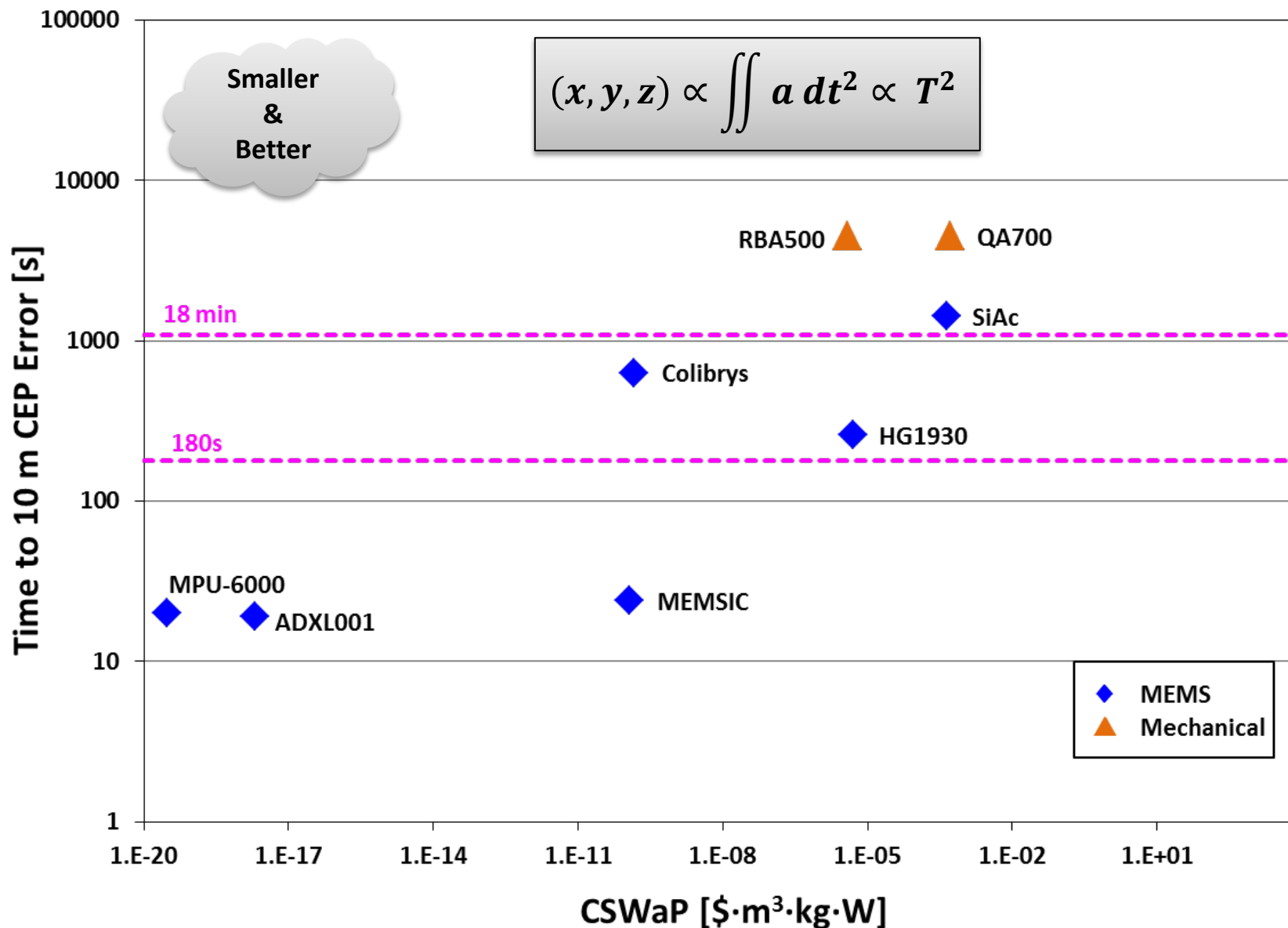
Simplified Missile/Munition Profiles





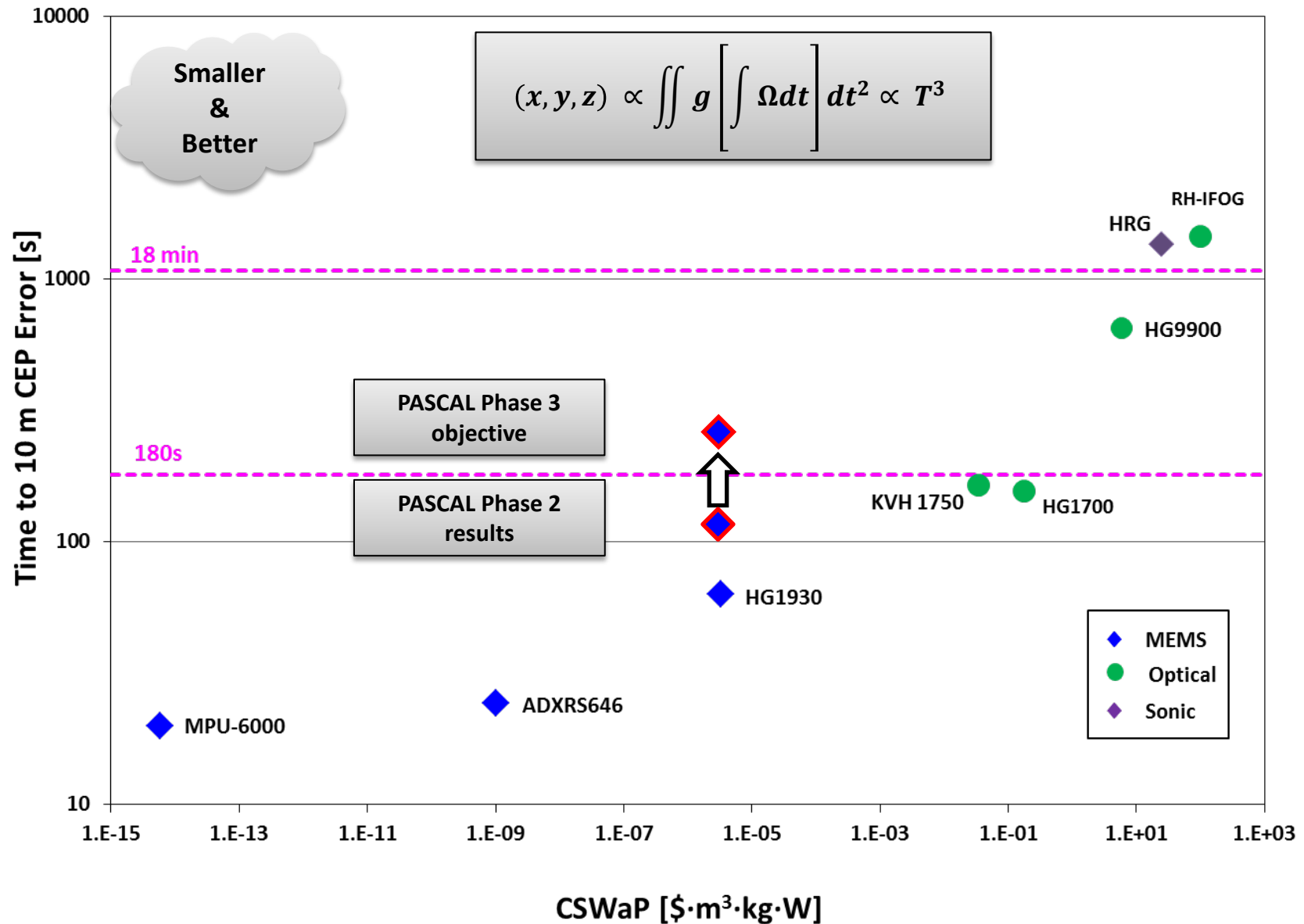
Munitions Navigation

	Current State	Capabilities Enabled by Navigation-Grade MEMS
Air-to-Ground Munition	<ul style="list-style-type: none">• Requires GPS tracking until impact• IMU used for stabilization• Inertial-only nav to 30 s (5 km) Challenge: Transfer alignment	<ul style="list-style-type: none">• Reduced CSWaP INS• Inertial-only nav to 180 s (60 km)• Reduce transfer alignment time
Ground-to-Ground Munition	<ul style="list-style-type: none">• Requires GPS tracking until impact• IMU used for stabilization• GPS required for terminal arming Challenge: Must survive gun launch	<ul style="list-style-type: none">• Inertial-only nav to 180 s (250 km)• Transfer alignment from GPS in first 10-15 s of glide flight• Must survive gun launch
Missile	<ul style="list-style-type: none">• IMU used for stabilization• Laser seeker for terminal guidance Challenges: Long-range navigation, transfer alignment	<ul style="list-style-type: none">• Extend terminal jamming radius to 180 s (250 km stand-off range)• Beyond line-of-sight navigation• Reduce transfer alignment time



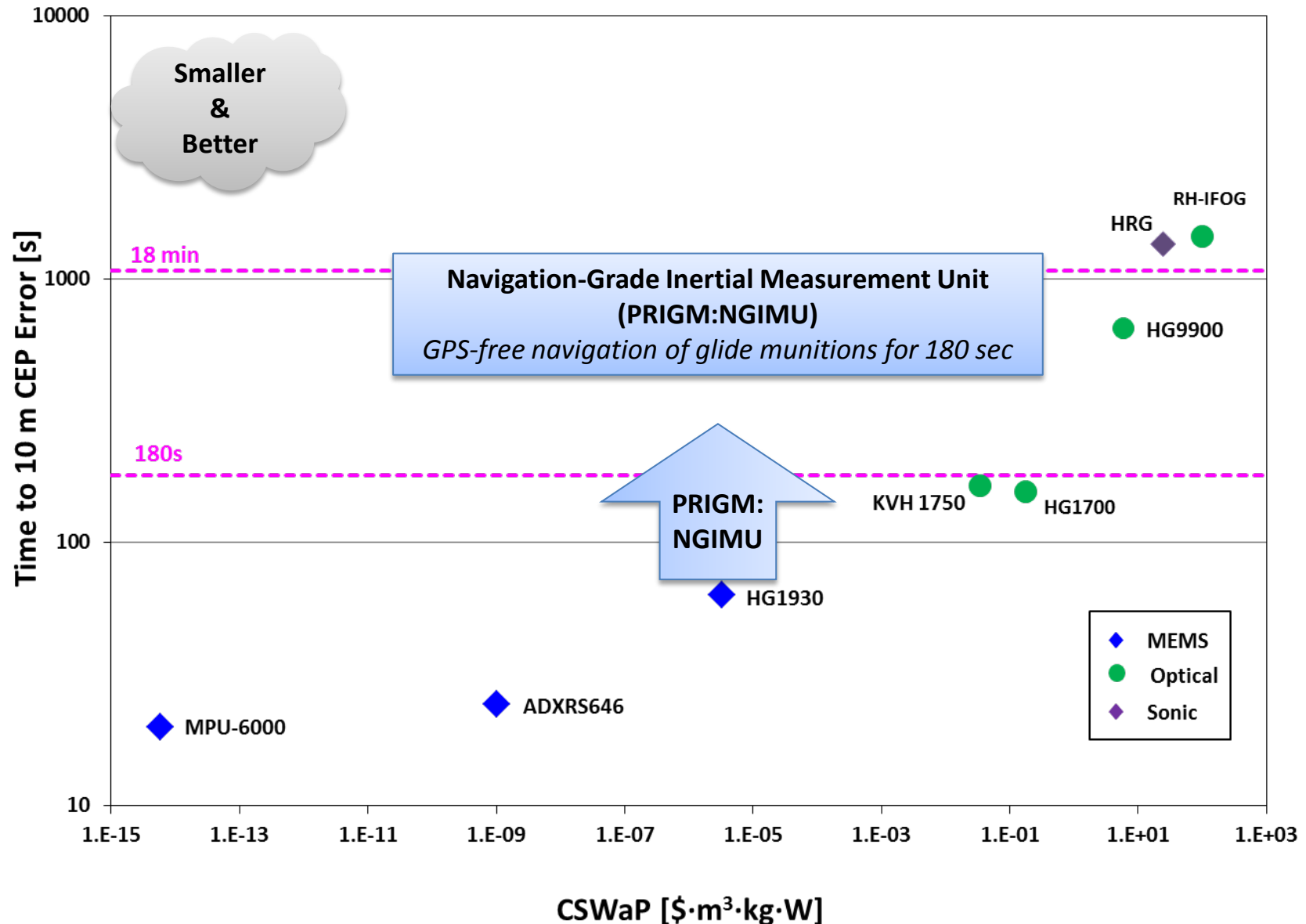


SOA Gyroscopes





SOA Gyros: Path Forward to Advanced MEMS IMUs





Nav-Grade Inertial Measurement Unit (PRIGM:NGIMU)

Motivation

- Enable guided munitions in GPS-contested theaters by 2020

Objective

- Eliminate compromise between **low-CSWaP, tactical-grade MEMS** and **high-CSWaP, navigation-grade RLG/iFOG-based IMUs**
- Develop prototype drop-in replacement **navigation-grade MEMS IMUs** in 2019
- Flight demos in 2020

Navigation-grade performance with MEMS CSWaP



© Honeywell

Nav-Grade IMU
Ring laser gyros, quartz accels



© Honeywell

Tactical-Grade IMU
MEMS gyros & accels



© Honeywell

Navigation-Grade IMU
SOA MEMS gyros & accels



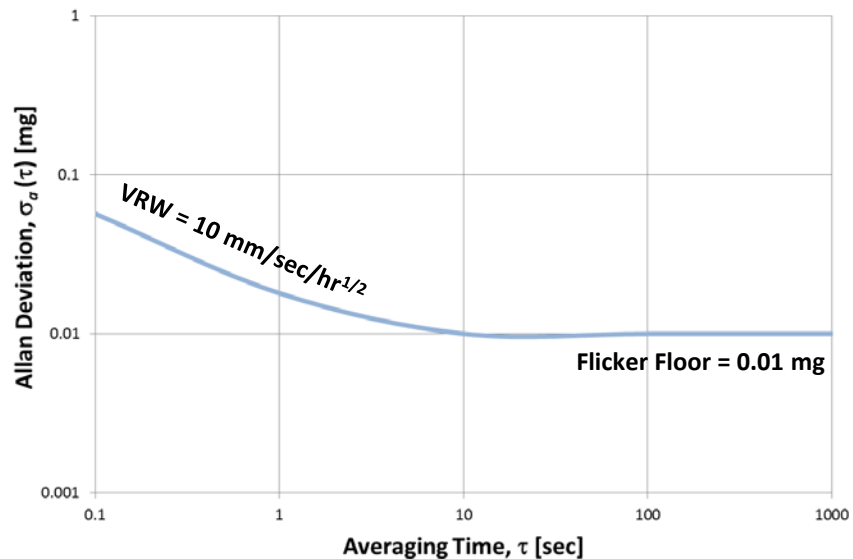
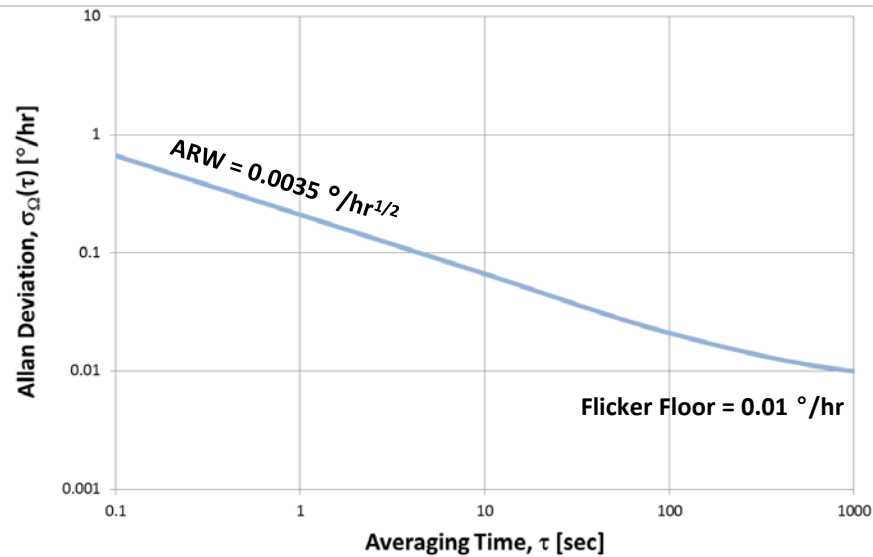
PRIGM:NGIMU Program Objectives

Program Deliverables: 10 MEMS-based IMUs with navigation-grade performance that are drop-in replacements for DoD-standard tactical-grade IMUs

Performance Metric	Objective	Units
Volume	82	cm ³
Weight	160	g
Power	< 3	W
Operating temperature range	-54 to +85	°C
Vibration DC to 2 kHz	7.7	g _{RMS}
Shock survivability	20,000	g
Bandwidth (min. @ -90° phase lag)	70	Hz
Gyroscope		
Operating range	± 900	°/sec
Turn-on to turn-on bias repeatability	0.01	°/hr, 1σ
Scale factor repeatability	5	ppm
Accelerometer		
Operating range	± 60	g
Turn-on to turn-on bias repeatability	25	μg, 1σ
Scale factor repeatability	25	ppm



PRIGM:NGIMU Program Objectives



Stability Specification (Allan Deviation)

τ [sec]	Gyroscope $\sigma_{\Omega}(\tau)$ [$^{\circ}/\text{hr}$]	Accelerometer $\sigma_a(\tau)$ [mg]
0.1	0.66	0.19
1	0.21	0.06
10	0.066	0.01
100	0.021	0.01
1000	0.01	0.01



Proposal Information

- A. Innovative Claims
- B. Technology Basis
- C. Detailed Technical Approach
- D. Test plan
- E. Risk analysis
- F. Schedule, milestones, and budget
- G. Technology Transfer
- H. Comparison with related research
- I. Biographies of key personnel
- J. Facilities
- K. Statement of Work (SOW)
- L. Summary Slide



Evaluation Criteria

- Overall scientific and technical merit
- Realism of proposed schedule
- Plans and capability to accomplish technology transition
- Cost realism
- Proposers' capabilities and/or related experience
- Potential contribution and relevance to the DARPA mission



www.darpa.mil